

CLARK Y	Re = 100000	Ma = 0.000	Ncrit = 9.000
CLARK Y AIRFOIL	Re = 200000	Ma = 0.000	Ncrit = 9.000
CLARK Y	Re = 400000	Ma = 0.000	Ncrit = 9.000
CLARK Y AIRFOIL	Re = 800000	Ma = 0.000	Ncrit = 9.000
CLARK Y AIRFOIL	Re = 70000	Ma = 0.000	Ncrit = 9.000
CLARK Y AIRFOIL	Re = 50000	Ma = 0.000	Ncrit = 9.000
CLARK Y AIRFOIL	Re $\sqrt{CL}$ = 100000	Ma $\sqrt{CL}$ = 0.000	Ncrit = 9.000

XFoil V 6.96

